

Issued	2001-11
Revised	2004-11
Reaffirmed	2012-04
Superseding AMS4330A	

Aluminum Alloy, Plate (2297-T87)  
2.8Cu - 1.5Li - 0.30Mn - 0.12Zr  
Solution Heat Treated, Stretched, and Artificially Aged  
(Composition similar to UNS A92297)

**RATIONALE**

AMS4330B has been reaffirmed to comply with the SAE five-year review policy.

**1. SCOPE:**

**1.1 Form:**

This specification covers an aluminum alloy in the form of plate.

**1.2 Application:**

These products have been typically used in aerospace applications where low density is needed in combination with moderate strength, high fatigue resistance, good stress corrosion properties, and improved stiffness, but usage is not limited to such applications.

**2. APPLICABLE DOCUMENTS:**

The issue of the following documents in effect on the date of the purchase order forms a part of this specification to the extent specified herein. The supplier may work to a subsequent revision of a document unless a specific document issue is specified. When the referenced document has been cancelled and no superseding document has been specified, the last published issue of that document shall apply.

**2.1 SAE Publications:**

Available from SAE, 400 Commonwealth Drive, Warrendale, PA 15096-0001 or [www.sae.org](http://www.sae.org).

AMS 2355      Quality Assurance Sampling and Testing, Aluminum Alloys and Magnesium Alloys, Wrought Products, Except Forging Stock, and Rolled, Forged, or Flash Welded Rings

AMS 2772      Heat Treatment of Aluminum Alloy Raw Materials

AS1990      Aluminum Alloy Tempers

SAE Technical Standards Board Rules provide that: "This report is published by SAE to advance the state of technical and engineering sciences. The use of this report is entirely voluntary, and its applicability and suitability for any particular use, including any patent infringement arising therefrom, is the sole responsibility of the user."

SAE reviews each technical report at least every five years at which time it may be reaffirmed, revised, or cancelled. SAE invites your written comments and suggestions.

Copyright © 2012 SAE International

All rights reserved. No part of this publication may be reproduced, stored in a retrieval system or transmitted, in any form or by any means, electronic, mechanical, photocopying, recording, or otherwise, without the prior written permission of SAE.

**TO PLACE A DOCUMENT ORDER:**    Tel: 877-606-7323 (inside USA and Canada)  
Tel: +1 724-776-4970 (outside USA)  
Fax: 724-776-0790  
Email: [CustomerService@sae.org](mailto:CustomerService@sae.org)  
**SAE WEB ADDRESS:**                    <http://www.sae.org>

**SAE values your input. To provide feedback on this Technical Report, please visit <http://www.sae.org/technical/standards/AMS4330B>**

## 2.2 ASTM Publications:

Available from ASTM, 100 Barr Harbor Drive, P.O. Box C700, West Conshohocken, PA 19428-2959 or [www.astm.org](http://www.astm.org).

ASTM B 594	Ultrasonic Inspection of Aluminum-Alloy Wrought Products for Aerospace Applications.
ASTM B 645	Plane-Strain Fracture Toughness Testing of Aluminum Alloys
ASTM B 660	Packaging/Packing of Aluminum and Magnesium Products
ASTM B 666/B 666M	Identification Marking of Aluminum and Magnesium Products
ASTM E 399	Plane-Strain Fracture Toughness of Metallic Materials
ASTM G 34	Exfoliation Corrosion Susceptibility in 2XXX and 7XXX Series Aluminum Alloys (EXCO Test)
ASTM G 47	Determining Susceptibility to Stress-Corrosion Cracking of 2XXX and 7XXX Aluminum Alloy Products

## 2.3 ANSI Publications:

Available from ANSI, 25 West 43<sup>rd</sup> Street, New York, NY 10036, or [www.ansi.org](http://www.ansi.org).

ANSI H35.2	Dimensional Tolerances for Aluminum Mill Products
ANSI H35.2M	Dimensional Tolerances for Aluminum Mill Products (Metric)

## 3. TECHNICAL REQUIREMENTS:

### 3.1 Composition:

Shall conform to the percentages by weight shown in Table 1, determined in accordance with AMS 2355.

Table 1 - Composition

Element	min	max
Silicon	--	0.10
Iron	--	0.10
Copper	2.5	3.1
Manganese	0.10	0.50
Magnesium	--	0.25
Zinc	--	0.05
Titanium	--	0.12
Lithium	1.1	1.7
Zirconium	0.08	0.15
Other Elements, each	--	0.05
Other Elements, total	--	0.15

### 3.2 Condition:

Solution heat treated at 980 to 1000 °F (536 to 538 °C) and rapidly cooled in a suitable quenching medium; stretched to produce a permanent set of at least 2.75% but not more than 6.75%; artificially aged to T87 temper (See AS 1990) at 320 °F (160 °C) for 20 to 48 hours. Aging time shall be dependent upon section thickness composition, and prior processing. Heat treatment shall be performed in accordance with AMS 2772.

3.2.1 Plate shall receive no further straightening operations after stretching.

### 3.3 Properties:

Plate shall conform to the following requirements, determined on the mill produced size in accordance with AMS 2355 and as specified herein.

3.3.1 Tensile Properties: Shall be as shown in Table 2.

TABLE 2A - Minimum Tensile Properties, Inch/Pound Units

Temper	Thickness, Inclusive (inches)	Grain Direction	Ultimate Tensile Strength (ksi)	Yield Strength 0.2% Offset (ksi)	% Elongation in 2 inches or 4D (percent)
T87	1.500 to 2.000	L	64.0	58.0	10
		LT	66.0	60.0	8
		ST	65.0	57.0	2
	2.001 to 2.500	L	63.0	57.0	9
		LT	65.0	59.0	7
		ST	64.0	56.0	2
	2.501 to 3.000	L	62.0	57.0	9
		LT	64.0	58.0	7
		ST	62.0	55.0	2
	3.001 to 4.000	L	62.0	57.0	5
		LT	62.0	57.0	4
		ST	59.0	54.0	1.5
	4.001 to 5.000	L	61.0	56.0	5
		LT	61.0	56.0	4
		ST	58.0	52.0	1.5
	5.001 to 6.000	L	60.0	55.0	5
		LT	60.0	55.0	4
		ST	57.0	52.0	1.5

TABLE 2B - Minimum Tensile Properties, SI Units

Temper	Thickness, Inclusive (mm)	Grain Direction	Ultimate Tensile Strength (MPa)	Yield Strength 0.2% Offset (MPa)	% Elongation in 50.8 mm or 4D %
	38.1 to 50.8	L	441	400	10
		LT	455	414	8
		ST	448	393	2
	50.9 to 63.5	L	434	393	9
		LT	448	407	7
		ST	441	386	2
	63.4 to 76.10	L	427	393	9
		LT	441	400	7
		ST	427	379	2
	76.20 to 101.60	L	427	393	5
		LT	427	393	4
		ST	407	372	1.5
	101.62 to 127.00	L	421	386	5
		LT	421	386	4
		ST	400	358	1.5
	127.02 to 152.40	L	414	379	5
		LT	414	379	4
		ST	393	358	1.5

## 3.3.2 Corrosion-Resistance:

- 3.3.2.1 Stress-Corrosion Resistance: Direct tension specimens machined and tested in accordance with ASTM G 47 shall show no evidence of stress corrosion failure when stressed in the short-transverse direction at 30.0 ksi (207 MPa) and exposed for 30 days.
- 3.3.2.2 Exfoliation Corrosion Resistance: When required by the purchaser, specimens from preproduction plates shall show exfoliation corrosion resistance equal to or better than EB when tested at the T/10 plane in accordance with ASTM G 34.
- 3.3.2.3 Plate not meeting the requirements of 3.3.2.1 or 3.3.2.2 may be given additional precipitation heat treatment. After such treatment, if all specified properties are met, the plate is acceptable.
- 3.3.2.4 Fracture Toughness: Fracture toughness shall be determined in accordance with ASTM E 399 and ASTM B 645 and shall meet the requirements for  $K_{Ic}$  specified in Table 3. For L-T and T-L test directions, use specimens with 1.5 inch (38.1 mm) minimum thickness. For the S-L test direction, specimen thickness should be maximized to the nearest 1/4 inch (6.35 mm). L-T, T-L and S-L specimens shall be centered at T/2 for plate 3.000 inches (76.2 mm) and under in nominal thickness and centered at T/4 for plate over 3.000 inches (76.2 mm) in nominal thickness. If a valid  $K_{Ic}$  or meaningful  $K_Q$  cannot be obtained due to insufficient specimen thickness or crack length, then the fracture toughness ( $K_Q$ ) of the specimen will be acceptable for lot release if the following conditions are met.

- 3.3.2.4.1 The B dimension of the specimen tested shall be the maximum possible up to 2.5 inches (63.5 mm) for the given plate thickness.
- 3.3.2.4.2 The specimen centerline location shall be maintained at the specified plate thickness location.
- 3.3.2.4.3 All fracture toughness validity checks except specimen thickness, crack length or  $P_{max}/P_Q$  shall meet the criteria of ASTM E 399 or ASTM B 645.
- 3.3.2.4.4 If the only invalidity with  $B=1.5$  inches (38.1 mm) is  $P_{max}/P_Q$ , test result is acceptable for lot release.

TABLE 3A - Minimum Fracture Parameters, Inch/Pound Units

Temper	Thickness, Inclusive (inches)	Specimen Orientation	Required Minimum Plane-Strain Fracture Toughness $K_{Ic}$ (ksi-in <sup>1/2</sup> )
T87	1.500 to 3.000	L-T	32
		T-L	27
		S-L	20
	3.001 to 4.000	L-T	31
		T-L	27
		S-L	20
	4.001 to 5.000	L-T	30
		T-L	26
		S-L	18
	5.001 to 6.000	L-T	29
		T-L	25
		S-L	18

TABLE 3B - Minimum Fracture Parameters, SI Units

Temper	Thickness, Inclusive (mm)	Specimen Orientation	Required Minimum Plane- Strain Fracture Toughness $K_{Ic}$ (MPa-m <sup>1/2</sup> )
T87	38.10 to 76.10	L-T	35.2
		T-L	29.7
		S-L	22.0
	76.20 to 101.60	L-T	34.1
		T-L	29.7
		S-L	22.0
	101.62 to 127.00	L-T	33.0
		T-L	28.6
		S-L	19.8
	127.02 to 152.40	L-T	31.9
		T-L	27.5
		S-L	19.8

#### 3.4 Quality:

Plate, as received by purchaser, shall be uniform in quality and condition, sound, and free from foreign materials and from imperfections detrimental to usage of the plate. Light scratches and discoloration or streaking shall not be reason for rejection.

- 3.4.1 Plate shall be ultrasonically inspected in accordance with ASTM B 594 and shall meet the requirements of Class A.

#### 3.5 Tolerances:

Shall conform to all applicable requirements of ANSI H35.2 or ANSI H35.2M.

#### 4. QUALITY ASSURANCE PROVISIONS:

##### 4.1 Responsibility for Inspection:

The vendor of plate shall supply all samples for vendor's tests and shall be responsible for the performance of all required tests. Purchaser reserves the right to sample and to perform any confirmatory testing deemed necessary to ensure that the plate conforms to specified requirements.

##### 4.2 Classification of Tests:

- 4.2.1 Acceptance Tests: Composition (3.1) is an acceptance test and shall be performed on each ingot or each group of ingots poured simultaneously from the same source of molten metal. Tensile properties (3.3.1), fracture toughness (3.3.2.4), and ultrasonic inspection (3.4.1) are acceptance tests and shall be performed on each parent plate.